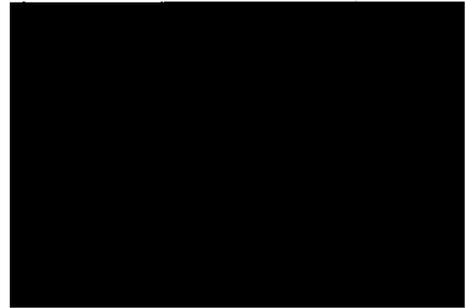


airfoil using PROFOIL in a few seconds and complete the com-



— A
- - - B

$Re = 9 \times 10^6$



upper surface to achieve a desired maximum thickness, while si-



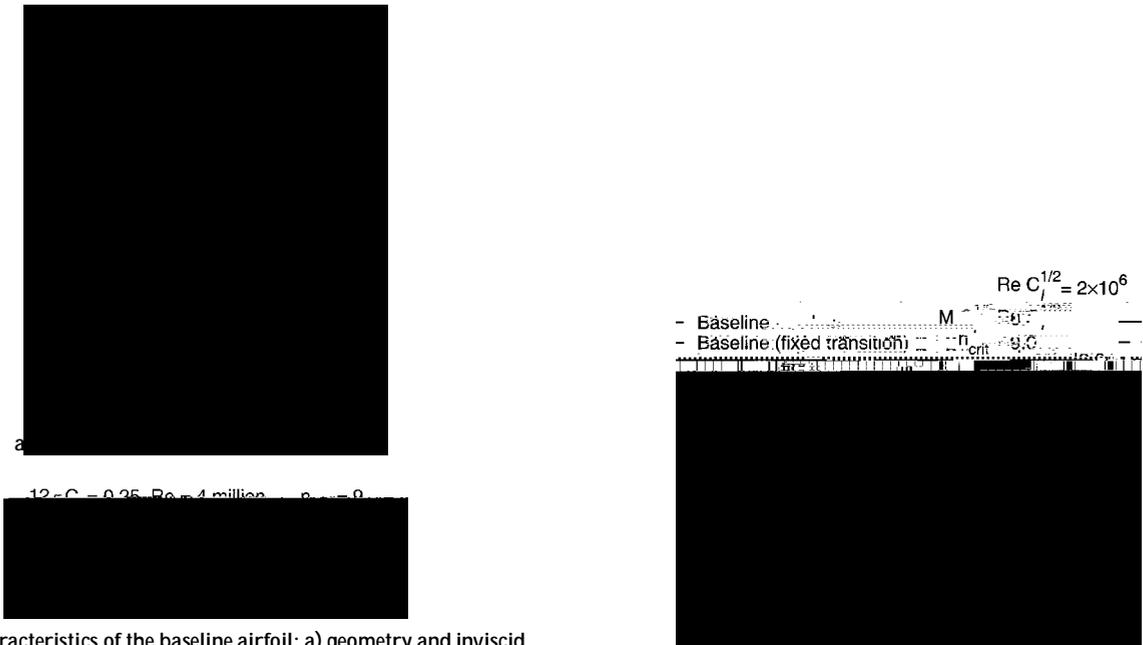
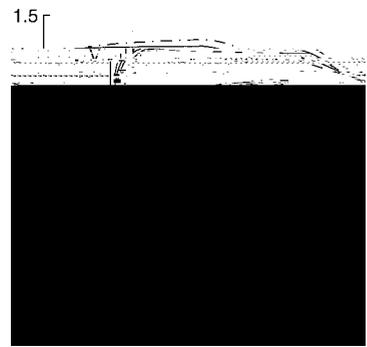
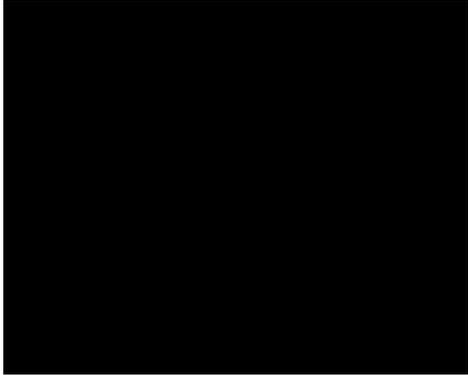
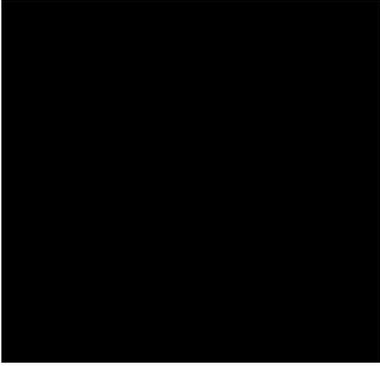


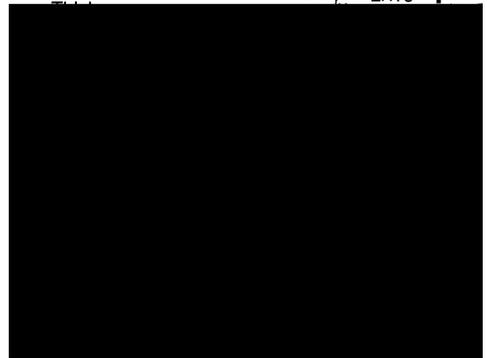
Fig. 3 Characteristics of the baseline airfoil: a) geometry and inviscid velocity distributions for the baseline airfoil (with important segment endpoints labeled) and b) development on the lower surface as computed by PROFOIL compared with the prescription.

C_l corresponding to the upper corner of the low-drag range to the leading edge at $C_{l,max}$.

By ensuring that the upper-surface transition point occurs naturally at the leading edge at stall, the $C_{l,max}$ value is made independent of the achievement of a significant amount of lift.



Re $C_{11}'' = 2 \times 10^6$ |



at a C_l of 0.4. Examination of the velocity

